



[4910-13-P]

**DEPARTMENT OF TRANSPORTATION**

**Federal Aviation Administration**

**14 CFR Part 39**

**[Docket No. FAA-2016-7271; Directorate Identifier 2015-NM-099-AD; Amendment 39-18722; AD 2016-24-05]**

**RIN 2120-AA64**

**Airworthiness Directives; Fokker Services B.V. Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for all Fokker Services B.V. Model F28 Mark 0070 and 0100 airplanes. This AD was prompted by heavy corrosion found on the wing rear spar lower girder. This AD requires inspections of the affected areas, modification of the wing trailing edge lower skin panels, and corrective actions if necessary. We are issuing this AD to address the unsafe condition on these products.

**DATES:** This AD is effective [INSERT DATE 35 DAYS AFTER DATE OF PUBLICATION IN THE FEDERAL REGISTER].

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of [INSERT DATE 35 DAYS AFTER DATE OF PUBLICATION IN THE FEDERAL REGISTER].

**ADDRESSES:** For service information identified in this final rule, contact, Fokker Services B.V., Technical Services Dept., P.O. Box 1357, 2130 EL Hoofddorp, the Netherlands; telephone +31 (0)88-6280-350; fax +31 (0)88-6280-111; email [technicalservices@fokker.com](mailto:technicalservices@fokker.com); Internet <http://www.myfokkerfleet.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-7271.

**Examining the AD Docket**

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-7271; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (telephone 800-647-5527) is Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

**FOR FURTHER INFORMATION CONTACT:** Tom Rodriguez, Aerospace Engineer, International Branch, ANM 116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone 425-227-1137; fax 425-227-1149.

## **SUPPLEMENTARY INFORMATION:**

### **Discussion**

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 by adding an AD that would apply to all Fokker Services B.V. Model F28 Mark 0070 and 0100 airplanes. The NPRM published in the Federal Register on June 23, 2016 (81 FR 40823) (“the NPRM”).

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Union, has issued EASA Airworthiness Directive 2015-0113, dated June 22, 2015 (referred to after this as the Mandatory Continuing Airworthiness Information, or “the MCAI”), to correct an unsafe condition for all Fokker Services B.V. Model F28 Mark 0070 and 0100 airplanes. The MCAI states:

On an F28 Mark 0070 aeroplane, heavy corrosion was found on the wing rear spar lower girder. At small spots the effective thickness of the vertical flange of the lower girder was almost lost. Subsequently, a number of inspections were accomplished on other aeroplanes to provide additional information on possible corrosion in this area. Because the rear spar lower girder between Wing Stations (WSTA) 9270 and 11794 is hidden from view by the inboard and outboard aileron balancing plates, it is possible that corrosion in this area remains undetected during the zonal inspections in zone 536 and 636 (MRB [Maintenance Review Board] tasks 062505—00-01 and 062605-00-01). The heavy corrosion was not only found in the area between WSTA 9270 and 11794, but also in the area where the rear spar lower girder is directly visible.

This condition, if not detected and corrected, reduces the load carrying capability of the wing, possibly resulting in structural failure and loss of the aeroplane.

To address this potential unsafe condition, Fokker Services issued Service Bulletin (SB) SBF100-57-049 to provide

instructions to detect and remove corrosion and to modify the wing trailing edge lower skin panels into access panels. SBF100-57-050 was issued to provide repair instructions.

For the reasons described above, this [EASA] AD requires inspections of the affected areas and, depending on findings, accomplishment of applicable corrective action(s) [including removing corrosion, repair, and restoring protective finish]. This [EASA] AD also requires modification of the wing trailing edge lower skin panels into access panels [This modification is to provide ease of access for later inspection and repairs in the affected areas.], and reporting of the results of the inspections to Fokker Services.

More information on this subject can be found in Fokker Services All Operators Message AOF100.197.

You may examine the MCAI in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-7271.

### **Comments**

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM or on the determination of the cost to the public.

### **Conclusion**

We reviewed the relevant data and determined that air safety and the public interest require adopting this AD as proposed except for minor editorial changes. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM.

## Related Service Information under 1 CFR part 51

We reviewed Fokker Service Bulletin SBF100-57-049, dated March 24, 2015, which describes procedures for an inspection for corrosion of certain wing rear spar lower girder areas, modification of the wing trailing edge lower skin panels, and corrective actions if necessary. We also reviewed Fokker Service Bulletin SBF100-57-050, Revision 1, dated May 19, 2015, which describes procedures for repair of the wing spar. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the ADDRESSES section.

### Costs of Compliance

We estimate that this AD affects 8 airplanes of U.S. registry.

We estimate the following costs to comply with this AD:

Estimated costs				
Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
Wing inspection and modification	35 work-hours X \$85 per hour = \$2,975 per inspection cycle	\$1,680	\$4,655 per inspection cycle	\$37,240 per inspection cycle
Reporting	1 work hour X \$85 per hour = \$85	\$0	\$85	\$680

We estimate the following costs to do any necessary corrective actions that will be required based on the results of the required inspection. We have no way of determining the number of airplanes that might need these corrective actions:

### On-condition costs

Action	Labor cost	Parts cost	Cost per product
Corrective Actions	Up to 372 work hours X \$85 per hour = \$31,620	Up to \$7,600	Up to \$39,220

### Paperwork Reduction Act

A federal agency may not conduct or sponsor, and a person is not required to respond to, nor shall a person be subject to penalty for failure to comply with a collection of information subject to the requirements of the Paperwork Reduction Act unless that collection of information displays a current valid OMB control number. The control number for the collection of information required by this AD is 2120-0056. The paperwork cost associated with this AD has been detailed in the Costs of Compliance section of this document and includes time for reviewing instructions, as well as completing and reviewing the collection of information. Therefore, all reporting associated with this AD is mandatory. Comments concerning the accuracy of this burden and suggestions for reducing the burden should be directed to the FAA at 800 Independence Ave., SW, Washington, DC 20591, ATTN: Information Collection Clearance Officer, AES-200.

### Authority for this Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in “Subtitle VII, Part A, Subpart III, Section 44701: General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

### **Regulatory Findings**

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

1. Is not a “significant regulatory action” under Executive Order 12866;
2. Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);
3. Will not affect intrastate aviation in Alaska; and
4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

### **List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

## **Adoption of the Amendment**

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

### **PART 39 - AIRWORTHINESS DIRECTIVES**

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

#### **§ 39.13 [Amended]**

2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

**2016-24-05 Fokker Services B.V.:** Amendment 39-18722; Docket No. FAA-2016-7271; Directorate Identifier 2015-NM-099-AD.

#### **(a) Effective Date**

This AD is effective [INSERT DATE 35 DAYS AFTER DATE OF PUBLICATION IN THE FEDERAL REGISTER].

#### **(b) Affected ADs**

None.

#### **(c) Applicability**

This AD applies to all Fokker Services B.V. Model F28 Mark 0070 and 0100 airplanes, certificated in any category.

#### **(d) Subject**

Air Transport Association (ATA) of America Code 57, Wings.

**(e) Reason**

This AD was prompted by heavy corrosion found on the wing rear spar lower girder. We are issuing this AD to detect and correct corrosion of the wing rear spar lower girder. This condition could reduce the load-carrying capability of the wing, possibly resulting in structural failure and loss of the airplane.

**(f) Compliance**

Comply with this AD within the compliance times specified, unless already done.

**(g) Inspection of the Wing Rear Spar Lower Girder from Wing Stations (WSTA) 9270 to 11794**

Within 1,000 flight cycles or 12 months, whichever occurs first after the effective date of this AD, accomplish a one-time detailed visual inspection for corrosion of the wing rear spar lower girder area from WSTA 9270 to 11794, in accordance with Part 1 of the Accomplishment Instructions of Fokker Service Bulletin SBF100-57-049, dated March 24, 2015.

**(h) Modification of Wing Trailing Edge**

Within 1,000 flight cycles or 12 months, whichever occurs first after the effective date of this AD, modify the wing trailing edge lower skin panels into access panels, in accordance with Part 1 of the Accomplishment Instructions of Fokker Service Bulletin SBF100-57-049, dated March 24, 2015.

**(i) Inspection of the Wing Rear Spar Lower Girder from WSTA 2635 to 8700 and WSTA 11794 to 12975**

Within 2,000 flight cycles or 24 months, whichever occurs first after the effective date of this AD, accomplish a one-time detailed visual inspection for corrosion of the

wing rear spar lower girder area from WSTA 2635 to 8700 and WSTA 11794 to 12975, in accordance with Part 2 of the Accomplishment Instructions of Fokker Service Bulletin SBF100-57-049, dated March 24, 2015.

**(j) Corrective Actions for the Inspections of Wing Rear Spar Lower Girder**

(1) If during any inspection required by paragraph (g) or (i) of this AD, as applicable, corrosion is found, before further flight, remove the corrosion and determine the remaining thickness at the damaged spots, in accordance with the Accomplishment Instructions of Fokker Service Bulletin SBF100-57-049, dated March 24, 2015. If the remaining thickness at the damaged spots, as determined by this paragraph, is not within the tolerances specified in Fokker Service Bulletin SBF100-57-049, dated March 24, 2015, except as required by paragraph (k)(1) of this AD: Before further flight, accomplish the applicable corrective actions as defined in paragraph (j)(1)(i) or (j)(1)(ii) of this AD, as applicable.

(i) For corrosion damage found outboard of WSTA 8200 only: Repair, in accordance with the Accomplishment Instructions of Fokker Service Bulletin SBF100-57-050, Revision 1, dated May 19, 2015.

(ii) Repair using a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or the European Aviation Safety Agency (EASA); or Fokker Services B.V.'s EASA Design Organization Approval (DOA).

(2) If during any inspection required by paragraph (g) or (i) of this AD, only damage to the surface protection is found, or if the remaining thickness at the damaged

spots, as determined by paragraph (j)(1) of this AD, is within the tolerances specified in Fokker Service Bulletin SBF100-57-049, dated March 24, 2015, except as required by paragraph (k)(1) of this AD: Before further flight, restore the surface protection, in accordance with the Accomplishment Instructions of Fokker Service Bulletin SBF100-57-049, dated March 24, 2015, except as required by paragraph (k)(2) of this AD.

**(k) Exceptions to Service Information Specifications**

(1) Where Fokker Service Bulletin SBF100-57-049, dated March 24, 2015, specifies the acceptability of smaller thickness or customized repairs: Before further flight, obtain acceptable tolerances, using a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or EASA; or Fokker Services B.V.'s EASA DOA.

(2) Where Fokker Service Bulletin SBF100-57-049, dated March 24, 2015, specifies contacting Fokker for a customized repair: Before further flight, repair using a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or EASA; or Fokker Services B.V.'s EASA DOA.

**(l) Reporting Requirements**

Submit a report of the findings, both positive and negative, of the inspections required by paragraphs (g) and (i) of this AD to Fokker Services, in accordance with the Accomplishment Instructions of Fokker Service Bulletin SBF100-57-049, dated March 24, 2015, at the time specified in paragraph (l)(1) or (l)(2) of this AD.

(1) If the inspection was done on or after the effective date of this AD: Submit the report within 30 days after the inspection.

(2) If the inspection was done before the effective date of this AD: Submit the report within 30 days after the effective date of this AD.

**(m) Other FAA AD Provisions**

The following provisions also apply to this AD:

**(1) Alternative Methods of Compliance (AMOCs):** The Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Branch, send it to ATTN: Tom Rodriguez, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone 425-227-1137; fax 425-227-1149. Information may be emailed to: 9-ANM-116-AMOC-REQUESTS@faa.gov. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

**(2) Contacting the Manufacturer:** For any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Branch, ANM-116, Transport Airplane

Directorate, FAA; or the EASA; or Fokker Service B.V.'s EASA DOA. If approved by the DOA, the approval must include the DOA-authorized signature.

**(3) Reporting Requirements:** A federal agency may not conduct or sponsor, and a person is not required to respond to, nor shall a person be subject to a penalty for failure to comply with a collection of information subject to the requirements of the Paperwork Reduction Act unless that collection of information displays a current valid OMB Control Number. The OMB Control Number for this information collection is 2120-0056. Public reporting for this collection of information is estimated to be approximately 5 minutes per response, including the time for reviewing instructions, completing and reviewing the collection of information. All responses to this collection of information are mandatory. Comments concerning the accuracy of this burden and suggestions for reducing the burden should be directed to the FAA at: 800 Independence Ave. SW, Washington, DC 20591, Attn: Information Collection Clearance Officer, AES-200.

**(n) Related Information**

Refer to Mandatory Continuing Airworthiness Information (MCAI) EASA Airworthiness Directive 2015-0113, dated June 22, 2015, for related information. This MCAI may be found in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-7271.

**(o) Material Incorporated by Reference**

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless this AD specifies otherwise.

(i) Fokker Service Bulletin SBF100-57-049, dated March 24, 2015.

(ii) Fokker Service Bulletin SBF100-57-050, Revision 1, dated May 19, 2015.

(3) For service information identified in this AD, contact Fokker Services B.V., Technical Services Dept., P.O. Box 1357, 2130 EL Hoofddorp, the Netherlands; telephone +31 (0)88-6280-350; fax +31 (0)88-6280-111; email [technicalservices@fokker.com](mailto:technicalservices@fokker.com); Internet <http://www.myfokkerfleet.com>.

(4) You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on November 17, 2016.

Phil Forde,  
Acting Manager,  
Transport Airplane Directorate,  
Aircraft Certification Service.

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